


EASA	EMERGENCY AIRWORTHINESS DIRECTIVE
	<p>AD No.: 2008-0129-E</p> <p>Date: 10 July 2008</p> <p>Note: This Emergency Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation</p>
This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].	
Type Approval Holder's Name : EADS SOCATA	Type/Model designation(s) : TBM 700 N
TCDS Number : EASA A.010	
Foreign AD : Not Applicable	
Supersedure : This Airworthiness Directive (AD) supersedes AD 2008-0067-E dated 03 April 2008	
ATA 21	Air Conditioning - Alternator and vapour cycle cooling system compressor support and drive assembly - Removal / Replacement / Inspection
Manufacturer(s):	EADS SOCATA
Applicability:	TBM 700 N aircraft from serial number (s/n) 434 and up. NOTE: TBM 850 is the commercial designation of the TBM700N.
Reason:	<p>Following the rupture of an alternator and vapour cycle cooling system pulley drive assembly, the AD 2008-0067-E had been published to require the replacement of the pulley drive assembly by a new one of an improved design.</p> <p>Recent cases of rupture of the alternator and vapour cycle cooling system compressor drive shaft and of cracks on the standby-alternator and compressor support have reportedly been found.</p> <p>Such failures could lead to the loss of the alternator and of the vapour cycle cooling systems, and could also cause mechanical damage inside the powerplant compartment.</p> <p>To address this condition, this AD supersedes AD 2008-0067-E and mandates the removal, as a temporary measure, of the compressor drive belt and of the torque limiter, the conditional replacement of the pulley drive shear shaft, and repetitive inspections for cracks of the pulley drive assembly and of the alternator/compressor support.</p>
Effective Date:	14 July 2008

<p>Required action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously.</p> <p>(1) For aircraft s/n 434 to 459 inclusive, prior to next flight after the effective date of this AD:</p> <ul style="list-style-type: none"> (a) Remove the pulley drive assembly, the torque limiter, the compressor drive belt and the alternator/compressor support, in accordance with paragraph (§) C of the accomplishment instructions of EADS SOCATA SERVICE BULLETIN (SB) 70-161-21 amendment 2 and (b) Inspect for cracks, the pulley drive surfaces and the alternator/compressor support welds, in accordance with § D.1 of the accomplishment instructions of SB 70-161-21 amendment 2. <p>If any crack is detected, before further flight, replace or conditionally repair the cracked unit as instructed in § D.2 of the accomplishment instructions of SB 70-161-21 amendment 2.</p> <ul style="list-style-type: none"> (c) Before accomplishment of § (d) of this AD, replace any pulley drive shear shaft that has never been replaced before the effective date of this AD, or that has, on the effective date of this AD, accumulated 30 Hours Time in Service (TIS) or more, with a serviceable shaft, in accordance with § E of the accomplishment instructions of SB 70-161-21 amendment 2. <p>NOTE 1: In the event an operator is unable to establish the accumulated hours TIS on a given shaft installed on an airplane, the total hours accumulated on the airplane must be used in the determination of the replacement time for the shaft.</p> <p>(d) Prior to next flight and until further notice:</p> <ul style="list-style-type: none"> - Re-install the pulley drive assembly and the alternator/compressor support, <u>without</u> re-installing compressor drive belt or the torque limiter, as instructed in § F of the accomplishment instructions of SB 70-161-21 amendment 2 and - Install on the instrument panel and in the Pilot's primary field of vision, the following placard <div style="border: 1px solid black; padding: 5px; text-align: center; margin: 10px auto; width: fit-content;"> <p>"AIR COND" INOPERATIVE RECOMMENDED "AIR COND" SWITCH POSITION: "MANUAL"</p> </div> <ul style="list-style-type: none"> - And insert the SB n° 70-161-21 amendment 2 in the Pilot's Operating Handbook. <p>Permission to ferry an airplane to a maintenance location to accomplish actions required by paragraph (1) of this AD is granted provided that the air conditioning is switched off during the all flight duration.</p> <p>(2) For all aircraft, at the next scheduled maintenance inspection or within 100 Flight Hours (FH) after the effective date of this AD, whichever occurs first, and thereafter at intervals not to exceed 100 FH, inspect for cracks, the pulley drive surfaces and the alternator/compressor support welds, in accordance with § C, D, F and G.1) to G.3) of the accomplishment instructions of SB 70-161-21 amendment 2.</p> <p>NOTE 2: For accomplishment of the repetitive inspections required by paragraph (2) of this AD, the paragraph C.2) of the accomplishment instructions of SB 70-161-21 amendment 2. does not apply since the torque limiter has already been removed.</p>
---	--

Ref. Publications:	<p>EADS SOCATA Service Bulletin 70-161-21 amendment 2.</p> <p>The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.</p>
Remarks :	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this EAD. 2. The safety assessment has requested not to implement the full consultation process and an immediate publication and notification. 3. Enquiries regarding this EAD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA. E-mail: ADs@easa.europa.eu. 4. For any questions concerning the technical content of the requirements in this EAD, please contact: <p style="text-align: center;">EADS SOCATA – Direction des Services – 65921 Tarbes Cedex 9 – France Tel. : +33 (0)5 62 41 73 00 Fax : +33 (0)5 62 41 76 54 or for the U.S.A SOCATA AIRCRAFT, INC. – North Perry Airport – 7501 South Airport Rd. Pembroke Pines, FL 33023 – United States of America Tel.: 1 (954) 893 1400 Fax: 1 (954) 964 4141</p>